Research Article

Vision-Based Nonlinear Control of Quadrotors Using the Photogrammetric Technique

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Received 12 June 2020; Revised 11 August 2020; Accepted 29 September 2020; Published 17 November 2020

Academic Editor: Aimé Lay-Ekuakille

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This paper presents a controller designed via the backstepping technique, for the tracking of a reference trajectory obtained via the photogrammetric technique. The dynamic equations used to represent the motion of the quadrotor helicopter are based on the Newton–Euler model. The resulting quadrotor model has been divided into four subsystems for the altitude, longitudinal, lateral, and yaw motions. A control input is designed for each subsystem. Furthermore, the photogrammetric technique has been used to obtain the reference trajectory to be tracked. The performance and effectiveness of the proposed nonlinear controllers have been tested via numerical simulations using the Pixhawk Pilot Support Package developed for Matlab/Simulink.

1. Introduction

The technology of unmanned aerial vehicles (UAVs) is developing rapidly, based on the integration of many technologies in the mechanical structure, energy management, communication, control, etc, [1]. UAVs are widely used in many applications in civilian as well as in military contexts. For instance, UAVs provide great advantages in geographical information acquisition and safety purposes, as well as significant benefits when used for inspection and detection purposes. Furthermore, UAVs are becoming more and more accessible due to technological developments that enable their increasing use in different and broad application areas.

Different mathematical models can be used to design a controller for UAVs. In [2–4], a Newton–Euler model was presented. Furthermore, the studies in [5, 6] considered quaternions to describe the angular kinematics, whilst the study in [7] applied the Euler–Lagrange equations to obtain the whole quadrotor mathematical model. Regarding the control of quadrotors, many control techniques were proposed. The linear quadratic regulator control and proportional integral derivative control were proposed in [8, 9]. However, the stability of these methods cannot be guaranteed when the quadrotor moves away from its equilibrium configuration. Compared to linear control, the nonlinear control can ensure global stable flight for a quadrotor. Examples of nonlinear control techniques used to design a controller for a quadrotor are the sliding mode and the backstepping techniques [10–13]. The backstepping and adaptive control [14] were applied to the quadrotor flight control. In [15, 16], an observer-based adaptive fuzzy
backstepping controller was designed for trajectory tracking of a quadrotor subject to wind gusts and parametric uncertainties. In [17], a robust adaptive attitude tracking controller parameter through a network learning process which technique and using neural networks to optimize the controller parameters through a network learning process which is based on the control process.

To design vision-based controllers, researchers used different methods for the recognition of objects, trajectories, road lanes, etc. In [19], an INS/ vision-based autonomous landing system on stationary platforms was proposed, whereas in [20], mobile platforms were used for the landing of quadrotors. In [21], the position of a quadrotor was estimated by the detection of a set of concentric circles. In [22], a controller, based on computer vision techniques applied to helipad recognition, was proposed, in which the visual recognition of a black and white pattern of the helipad was exploited. In [23], a computer vision system was developed for the automatic estimation of the position and attitude of a helipad fixed on a mobile platform. In [24], a trajectory tracking problem for a vision-based quadrotor control system via super twisting sliding mode controller was proposed. Also, the study in [25] proposed a neuroadaptive integral robust controller for the tracking of ground moving targets in the presence of various uncertainties, via an Image-Based Visual Servo (IBVS) framework. The authors of [26] presented a predictive control algorithm for autonomous approaches of quadrotor helicopters to a window using a reference image captured with a photographic camera. The target is selected by an operator in a reference image which is sent to the vehicle. Besides, the authors of [27] proposed an adaptive sliding mode controller based on the backstepping technique for a tracking problem using a monocular algorithm to obtain the accurate location information of the quadrotor and its reference. In the same context, the work in [28, 29] developed a vision-based attitude dynamic surface controller, constructing an IBVS to ensure that the visual target remains in the camera’s field of view all the time.

In this paper, a controller for the tracking of a reference trajectory is designed via the backstepping technique. The quadrotor model is divided into four subsystems for the altitude, longitudinal, lateral, and yaw motions, and a control input is designed for each subsystem. Furthermore, the photogrammetric technique is used to obtain the reference trajectory to be tracked. The performance and effectiveness of the proposed nonlinear controllers are tested via numerical simulations using the simulation software called Pixhawk Pilot Support Package (PSP), which predicts accurately the real dynamic quadrotor helicopter behavior. The main contributions of this article are as follows:

(i) The backstepping technique is used to design a controller capable of tracking a reference position and yaw of a quadrotor

(ii) The photogrammetric technique is used to obtain the reference trajectory

(iii) The performance and effectiveness of the proposed controller have been tested in PSP

The paper is organized as follows: Section 2 introduces the description and the mathematical model of the quadrotor. In Section 3, the control problem is formulated, whilst in Section 4, the nonlinear controller is designed. In Section 5, the reference trajectory is obtained via the photogrammetric technique, and some numerical simulations implemented in PSP are presented. Some comments conclude the paper.

2. Mathematical Model of a Quadrotor

The quadrotor considered in this work consists of a rigid frame equipped with four rotors. The rotors generate the propeller force \( F_i = b \omega_{\text{prop},i} \), proportional to the propeller angular velocity \( \omega_{\text{prop},i} \), \( i = 1, 2, 3, 4 \). Propellers 1 and 3 rotate counterclockwise, and propellers 2 and 4 rotate clockwise.

Let us indicate with \( RC(O,e_1,e_2,e_3) \) and \( R\Gamma(\Omega,e_1,e_2,e_3) \) the frames fixed with the Earth and the quadrotor, respectively, and \( \Omega \) is coincident with the center of mass of the quadrotor (see Figure 1). The absolute position of quadrotor in RC is described by \( p = (x, y, z)^T \), whereas its attitude is described by the Euler angles \( \alpha = (\phi, \theta, \psi)^T \), with \( \phi, \theta, \psi \in (-\pi/2, \pi/2) \) being the pitch, roll, and yaw angles, respectively. The sequence 3–2–1 is here considered [30]. Moreover, \( v = (v_x, v_y, v_z)^T \) and \( \omega = (\omega_1, \omega_2, \omega_3)^T \) are the linear and angular velocities of the center of mass of the quadrotor, expressed in RC and in \( R\Gamma \), respectively.

The translation dynamics, expressed in RC, and rotation dynamics, expressed in \( R\Gamma \), of the quadrotor are

\[
\begin{align*}
\dot{p} &= v, \\
\dot{v} &= \frac{1}{m} \mathcal{R}(\alpha) F_{\text{prop}} + \frac{1}{m} F_{\text{grav}} + \frac{1}{m} F_d, \\
\dot{\alpha} &= M(\alpha) \omega, \\
\dot{\omega} &= J^{-1}(-\bar{\omega} I \omega + \tau_{\text{prop}} - \tau_{\text{gyro}} + M_d),
\end{align*}
\]

where \( m \) is the mass of the quadrotor, \( J = \text{diag}[J_x, J_y, J_z] \) (expressed in \( R\Gamma \)) is the inertia matrix of the quadrotor, and

\[
\dot{\bar{\omega}} = \begin{pmatrix}
0 & -\omega_3 & \omega_2 \\
\omega_3 & 0 & -\omega_1 \\
-\omega_2 & \omega_1 & 0
\end{pmatrix}.
\]
Considering small angles, matrix (5) can be approximated by the identity matrix, i.e., \( M(\alpha) = I_{3 \times 3} \). This assumption is justified by the fact that the movements of the quadrotor are slow and soft [4].

The rolling torque \( \tau_1 \) is produced by the forces \( F_2 \) and \( F_4 \). Similarly, the pitching torque \( \tau_3 \) is produced by the forces \( F_1 \) and \( F_3 \). Due to Newton’s third law, the propellers produce a yawing torque \( \tau_2 \) on the body of the quadrotor, in the opposite direction of the propeller rotation. Moreover,

\[
\tau_{\text{gyro}} = \sum_{i=1}^{4} (-1)^{i+1} J_p \omega_p \omega \epsilon_z, 
\]

is the gyroscopic torque due to the propeller rotations, with \( J_p \) being the propeller moment of inertia with respect to its rotation axis, and \( \omega_p = \omega_{p,3} - \omega_{p,2} + \omega_{p,1} - \omega_{p,4} \) is the so-called rotor relative speed. Finally, \( F_j \) and \( M_j \) are the forces and torques due to the external disturbances, here assumed negligible.

Now, using (1) and (3) and under the small angle assumption, the mathematical model of the quadrotor can be expressed by

\[
\begin{align*}
\dot{x} &= \frac{1}{m} (c_x s_\theta c_\psi + s_x s_\theta s_\psi) u_1, \\
\dot{y} &= \frac{1}{m} (c_x s_\theta c_\psi - s_x s_\theta s_\psi) u_1, \\
\dot{z} &= -g + \frac{1}{m} c_\theta c_\psi u_1, \\
\dot{\phi} &= \frac{J_y - J_z \phi - J_y \phi + 1}{J_y} \dot{u}_2, \\
\dot{\theta} &= \frac{J_z - J_x \phi - J_x \phi + 1}{J_x} \dot{u}_2, \\
\dot{\psi} &= \frac{J_z - J_y \phi - J_y \phi + 1}{J_z} \dot{u}_4, \\
\end{align*}
\]

where the control variables \( u_j, j = 1, 2, 3, 4 \), are defined as (3). The values of the parameters used in (7) are defined in Table 1.

3. Formulation of the State Feedback Control Problem

The control problem is to ensure the asymptotic converge of the variables \( \chi = (x, y, z, \psi) \) to some reference trajectories \( \chi_{\text{ref}} = (x_{\text{ref}}, y_{\text{ref}}, z_{\text{ref}}, \psi_{\text{ref}}) \). In view of the control of the quadrotor, the following coordinate change is considered: \( \phi = \omega_\phi, \theta = \omega_\theta, \) and \( \psi = \omega_\psi (\omega_\phi = \omega_1, \omega_\theta = \omega_2, \omega_\psi = \omega_3) \), and the system of equations (7) became
\[
\begin{align*}
\dot{x} &= v_x, \\
\dot{v}_x &= \frac{1}{m}(c_\phi s_\theta c_\psi + s_\phi s_\psi)u_1, \\
\dot{y} &= v_y, \\
\dot{v}_y &= \frac{1}{m}(c_\phi s_\theta s_\psi - s_\phi c_\psi)u_1, \\
\dot{z} &= v_z, \\
\dot{v}_z &= -g + \frac{1}{m}c_\phi c_\theta u_1, \\
\dot{\phi} &= \omega_\phi, \\
\dot{\omega}_\phi &= \frac{J_y - J_z}{J_x} \omega_\theta \omega_\psi - \frac{J_p}{J_x} \omega_\phi \omega_\theta + \frac{l}{J_x} \omega_3, \\
\dot{\theta} &= \omega_\theta, \\
\dot{\omega}_\theta &= \frac{J_z - J_x}{J_y} \omega_\phi \omega_\psi + \frac{J_p}{J_y} \omega_\phi \omega_\theta + \frac{l}{J_y} \omega_2, \\
\dot{\psi} &= \omega_\psi, \\
\dot{\omega}_\psi &= \frac{J_x - J_y}{J_z} \omega_\phi \omega_\theta + \frac{1}{J_z} \omega_4.
\end{align*}
\]

Clearly, model (8) is composed of rotational and translational dynamics. Figure 2 shows that the input control \(u_1\) does not influence the rotational dynamics but only the translational ones and can be used to impose \(z \rightarrow z_{\text{ref}}\). Furthermore, in the spirit of the backstepping technique, the \((\phi, \theta, \psi)\) can be used to impose \(x \rightarrow x_{\text{ref}}, y \rightarrow y_{\text{ref}}\). In turn, \((\phi, \theta, \psi)\) are influenced by the inputs \(u_1, u_2,\) and \(u_4\), which can be used to impose \(\phi \rightarrow \phi_{\text{ref}}, \theta \rightarrow \theta_{\text{ref}},\) and \(\psi \rightarrow \psi_{\text{ref}}\). Hence, the following four subsystems will be considered: the altitude subsystem \(S_1,\) the latitudinal subsystem \(S_2,\) the longitudinal subsystem \(S_3,\) and the yaw subsystem \(S_4:\)

\[
\begin{align*}
S_1 &= \begin{cases} 
\dot{z} &= v_z, \\
\dot{v}_z &= -g + \frac{1}{m}c_\phi c_\theta u_1, 
\end{cases} \\
S_2 &= \begin{cases} 
\dot{\phi} &= \omega_\phi, \\
\dot{\omega}_\phi &= \frac{J_y - J_z}{J_x} \omega_\theta \omega_\psi - \frac{J_p}{J_x} \omega_\phi \omega_\theta + \frac{l}{J_x} \omega_3, 
\end{cases} \\
S_3 &= \begin{cases} 
\dot{\theta} &= \omega_\theta, \\
\dot{\omega}_\theta &= \frac{J_z - J_x}{J_y} \omega_\phi \omega_\psi + \frac{J_p}{J_y} \omega_\phi \omega_\theta + \frac{l}{J_y} \omega_2, 
\end{cases} \\
S_4 &= \begin{cases} 
\dot{\psi} &= \omega_\psi, \\
\dot{\omega}_\psi &= \frac{J_x - J_y}{J_z} \omega_\phi \omega_\theta + \frac{1}{J_z} \omega_4.
\end{cases}
\end{align*}
\]
The controller design for each subsystem will be carried out in the following section.

4. Design of the Nonlinear Controller

The translation dynamics in equation (7) are dependent on the \( \phi, \theta, \) and \( \psi \) angles. In the next sections, a nonlinear control is used for each subsystem to solve the control problem.

4.1. Altitude Control. In this section are established the altitude control and its stability proof via Lyapunov function. For this purpose, it is considered that the altitude control \( S_1 \) subsystem is

\[
\ddot{z} = v_z, \quad \dot{v}_z = -g + \frac{1}{m} c_q c_\vartheta u_1.
\]

The control aim is to maintain the quadrotor at a desired constant altitude \( z_{\text{ref}} \), with \( v_z, \dot{z}_r = \dot{z}_{\text{ref}} \). To this aim, setting the tracking error for the altitude control as

\[
e_{z,1} = z - z_{\text{ref}}, \quad e_{z,2} = v_z - v_{z,\text{ref}} + k_{z,1} e_{z,1},
\]

the dynamic errors are

\[
\dot{e}_{z,1} = -k_{z,1} e_{z,1} + e_{z,2}, \quad \dot{e}_{z,2} = -g + \frac{1}{m} c_q c_\vartheta u_1 - \dot{v}_z_{\text{ref}} + k_{z,1} (e_{z,2} - k_{z,1} e_{z,1} + v_{z,\text{ref}} - \dot{z}_{\text{ref}}).
\]

The proposed altitude control \( u_1 \) is

\[
u_1 = \frac{m}{c_q c_\vartheta} (g + \dot{v}_z_{\text{ref}} - k_{z,1} (e_{z,2} - k_{z,1} e_{z,1} + v_{z,\text{ref}} - \dot{z}_{\text{ref}})) - k_3 e_{z,1} - k_4 e_{z,2},
\]

where \( c_\vartheta \neq 0 \) and \( c_q \neq 0 \). Therefore, using altitude control (13) in (12), the altitude error dynamics became

\[
\begin{pmatrix}
\dot{e}_{z,1} \\
\dot{e}_{z,2}
\end{pmatrix} = A_z
\begin{pmatrix}
e_{z,1} \\
e_{z,2}
\end{pmatrix}, \quad A_z = \begin{pmatrix}
-k_{z,1} & 1 \\
-k_3 & -k_4
\end{pmatrix}.
\]

For the stability analysis of system (14), the following Lyapunov function candidate is proposed:

\[
V_z (e_z) = \frac{1}{2} e_z^T P_z e_z,
\]

\[
e_z = \begin{pmatrix}
e_{z,1} \\
e_{z,2}
\end{pmatrix},
\]

\[
P_z = \begin{pmatrix}
p_{z,11} & p_{z,12} \\
p_{z,12} & p_{z,22}
\end{pmatrix},
\]

where \( P_z = P_z^T > 0 \). Deriving Lyapunov function (15), one obtains

\[
\dot{V}_z (e_z) = e_z^T (A_z^T P_z + P_z A_z) e_z,
\]

where \( P_z \) is taken as solution of the Lyapunov equation

\[
A_z^T P_z + P_z A_z = -Q_z,
\]

\[
Q_z = \begin{pmatrix}
q_{1,z} & 0 \\
0 & q_{1,z}
\end{pmatrix},
\]

with \( Q_z = Q_z^T > 0 \) being fixed. Using (17) into (16), one finally obtains

\[
\dot{V}_z (e_z) = -e_z^T Q_z e_z \leq \lambda_{\min} \| e_z \|^2 \leq -\alpha V_z (e_z),
\]

with \( \alpha = 2 \lambda_{\min} \| P_z \|_{\text{max}} \). Therefore, by \( [31] \), \( V_z (e_z) \leq e^{-\alpha t} V (e_z (0)) \), so that the tracking errors converge globally exponential to zero.

4.2. Longitudinal Motion Control. In this section, the longitudinal motion is studied making use of the subsystem \( S_2 \) and using the backstepping technique. One considers the tracking error \( e_{x,1} = x - x_{\text{ref}} \), where \( v_{x,\text{ref}} = \dot{x}_{\text{ref}} \), and the error dynamics

\[
\dot{e}_{x,1} = v_x - \dot{x}_{\text{ref}},
\]

or, equivalently,

\[
\dot{e}_{x,1} = -k_{x,1} e_{x,1} + (v_x - \dot{x}_{\text{ref}} + k_{x,1} e_{x,1})
\]

where

\[
e_{x,2} = v_x - v_{x,\text{ref}}, \quad k_{x,1} > 0,
\]

is the velocity error and

\[
v_{x,\text{ref}} = \dot{x}_{\text{ref}} - k_{x,1} e_{x,1},
\]
\[ \dot{e}_{\theta,2} = \frac{J_x - J_y}{J_y} \omega_x \omega_y + \frac{J_y}{J_y} \omega_y \omega_x + \frac{1}{J_y} u_2 - \dot{\theta}_{\text{ref}}. \]  

Finally, one determines the input control \( u_2 \) as

\[ u_2 = \frac{J_z - J_y}{J_y} \omega_x \omega_y - \frac{J_y}{J_z} \omega_y \omega_x - \frac{1}{J_z} k_{\theta,2} \dot{e}_{\theta,2}. \]  

so that the velocity error dynamics become

\[ \dot{e}_{\theta,2} = -k_{\theta,2} e_{\theta,2}, \]  

with \( k_{\theta,2} > 0 \).

### 4.3. Lateral Motion Control

In this section is designed the controller for the lateral motion along the \( y \)-axis of the quadrotor, using the same procedure followed in Section 4.2. One starts with the tracking error

\[ e_{y,1} = y - y_{\text{ref}}, \]  

and the dynamics

\[ \dot{e}_{y,1} = v_y - \dot{y}_{\text{ref}} = -k_{y,1} e_{y,1} + \left( v_y - \dot{y}_{\text{ref}} + k_{y,1} e_{y,1} \right), \]

\[ k_{y,1} > 0, \]

\[ e_{y,2} = v_y - v_{y,\text{ref}}, \]

where

\[ v_{y,\text{ref}} = \dot{y}_{\text{ref}} - k_{y,1} e_{y,1}. \]  

The dynamics of \( e_{y,2} \) are

\[ \dot{e}_{y,2} = \dot{v}_y - \dot{v}_{y,\text{ref}} = -k_{y,2} e_{y,2} - \frac{1}{m} c_y u_4, \]  

\[ s_{\phi,1} = \frac{m}{c_y} \left( \dot{y}_{\text{ref}} - k_{y,1} e_{y,1} + \left( k_{y,1} + k_{y,2} \right) e_{y,2} \right) + \frac{s}{c_y} c_y s_{\phi,1}, \]

\[ e_{\phi,1} = s_{\phi} - s_{\phi,\text{ref}}, \]

\[ \phi_{\text{ref}} = \arcsin s_{\phi,\text{ref}}, \]  

with \( k_{\phi,2} > 0, c_y \neq 0 \) and \( u_1 \) given by (13). Furthermore, the angular error dynamics are

\[ \dot{e}_{\phi,1} = -k_{\phi,1} e_{\phi,1} + c_y \omega_x - \dot{\phi}_{\text{ref}} e_{\phi,1} + k_{\phi,1} e_{\phi,1}, \]

\[ = -k_{\phi,1} e_{\phi,1} + c_y \omega_x, \]  

with \( k_{\phi,1} > 0 \) and

\[ e_{\phi,2} = \omega_x - \omega_{\phi,\text{ref}}, \]

\[ \omega_{\phi,\text{ref}} = \frac{1}{c_y} \left( \phi_{\text{ref}} - k_{\phi,1} e_{\phi,1} \right). \]  

Finally, the angular velocity error dynamics are

\[ \dot{e}_{\phi,2} = \frac{J_y - J_z}{J_y} \omega_y + \frac{J_y}{J_y} \omega_y + \frac{1}{J_y} \dot{u}_3 - \dot{\phi}_{\text{ref}}. \]  

so that the control input is

\[ u_3 = \frac{J_y - J_z}{J_y} \omega_y + \frac{J_y}{J_y} \omega_y + \frac{1}{J_y} \dot{u}_3 - \dot{\phi}_{\text{ref}}. \]  

with \( k_{\phi,2} > 0 \).

#### 4.4. Yaw Motion Control

Considering the subsystem \( S_4 \), and the errors

\[ e_{\psi,1} = \psi - \psi_{\text{ref}}, \]

\[ e_{\psi,2} = \omega_{\psi} - \omega_{\psi,\text{ref}}, \]  

one works out the error dynamics as

\[ \dot{e}_{\psi,1} = \omega_{\psi} - \psi_{\text{ref}} = -k_{\psi,1} e_{\psi,1} + e_{\psi,2}, \]

\[ \omega_{\psi,\text{ref}} = \psi_{\text{ref}} - k_{\psi,1} e_{\psi,1}, \]

\[ \hat{e}_{\psi,2} = \frac{J_x - J_y}{J_y} \omega_\theta - \frac{1}{J_y} \dot{u}_4 - \dot{\omega}_{\psi,\text{ref}}. \]

\[ \hat{\omega}_{\psi,\text{ref}} = \frac{1}{k_{\psi,1}} e_{\psi,1} - k_{\psi,1} e_{\psi,2}, \]
with \(k_{\psi,1} > 0\). Therefore, with the control
\[
\dot{u}_4 = -(J_x - J_y)\dot{\theta} + J_z \dot{\omega}_y + J_z \dot{\omega}_y - k_{\psi,2} e_{\psi,2}, \ k_{\psi,2} > 0,
\]
(41)
on one finally obtains the yaw error dynamics as
\[
\begin{pmatrix}
\dot{e}_{\psi,1} \\
\dot{e}_{\psi,2}
\end{pmatrix} =
\begin{pmatrix}
-k_{\psi,1} & 1 \\
0 & -k_{\psi,2}
\end{pmatrix}
\begin{pmatrix}
e_{\psi,1} \\
e_{\psi,2}
\end{pmatrix}.
\]
(42)

The global exponential stability can be inferred analogously to what is done in Section 4.1.

5. Simulation Results

The photogrammetric technique has been used for measuring the important physical quantities in both ground and flight testing, including attitude, position, and shape of objects. To generate the trajectory reference
\[
P_{\text{ref}}(t) = (x_{\text{ref}}(t), y_{\text{ref}}(t), z_{\text{ref}}(t)),
\]
a photogrammetric technique was applied to analyze a natural scenery, with
\[
\begin{align*}
x_{\text{ref}}(t) &= f_1(x(t), y(t), z(t)), \\
y_{\text{ref}}(t) &= f_2(x(t), y(t), z(t)), \\
z_{\text{ref}}(t) &= f_3(x(t), y(t), z(t)),
\end{align*}
\]
and with \(f_i, \ i = 1, 2, 3\), functions of \((x(t), y(t), z(t))\). The selected scenario is a serial array of trees, where all trees have approximately the same separation. Figures 3 and 4 show the trees’ frontal and lateral views, respectively. Using the photogrammetry, the following estimations were determined: \(W_f = 0.85\) m for the trees’ width, \(T_t = 2.72\) m for the distance between the two trees, and \(W_{s,1} = 0.75\) m and \(W_{s,2} = 0.522\) m for the widths of the two trees.

For this application, the reference trajectory has two important characteristics:

1. The altitude is constant
2. The reference trajectory is periodic over the \(x \sim y\) plane
3. The reference trajectory avoids the collision between the quadrotor and the trees

Therefore, the reference position was set equal to
\[
P_{\text{ref}}(t) = \begin{pmatrix} x_{\text{ref}}(t) \\ y_{\text{ref}}(t) \\ z_{\text{ref}}(t) \end{pmatrix} = \begin{pmatrix} v_{x,\text{ref}} t \\ A_{\text{ref}} \sin(2\pi f_{\text{ref}} x_{\text{ref}}(t)) \\ 0.5t \end{pmatrix},
\]
(45)
with \(v_{x,\text{ref}} = 0.5\) m/s, \(A_{\text{ref}}\) being the amplitude, and \(f_{\text{ref}}\) being the frequency of the periodic function, such as
\[
f_{\text{ref}} = \frac{1}{T_{\text{ref}}} = \frac{1}{2(W_{s,1} + W_{s,2} + T_t)} = 0.125\ \text{cycles/m},
\]
(46)
where \(T_{\text{ref}}\) is the period. Figure 5 shows the reference trajectory for \(x_{\text{ref}} \in [0, 18]\) m.

The behavior of the controllers designed in Section 4 has been tested with numerical simulations on the quadrotor described by equation (1). The parameters used are given in Table 1, whilst the variables and gains used in the controllers
are given in Table 2. In order to simulate the controller for each subsystem, the software developed by the PSP for Simulink was used. The reason for choosing the PSP software was due to its performance in predicting the dynamic quadcopter behavior, which is very close to the real behavior. In fact, the PSP software is proven to represent accurately the quadrotor dynamics. In order to appropriately implement the controllers, the reference trajectory was sampled as in Figure 6.

Figure 7 shows the quadrotor’s altitude $z(t)$ (solid) and reference altitude $z_{ref}(t)$ (dashed); (b) quadrotor’s yaw angle $\psi(t)$ (solid) and yaw angle reference $\psi_{ref}(t)$ (dashed).

Figure 8 shows the quadrotor's altitude $z(t)$ (solid) and reference altitude $z_{ref}(t)$ (dashed); (b) quadrotor's yaw angle $\psi(t)$ (solid) and yaw angle reference $\psi_{ref}(t)$ (dashed).

The simulations results of the closed-loop system are shown in Figures 7 and 8, where the quadrotor’s initial conditions have been set equal to $p(0) = (0, 0, 0)^T$, $v(0) = (0, 0, 0)^T$, $\alpha(0) = (0, 0, 0)^T$, and $\omega(0) = (0, 0, 0)^T$. Figure 7 shows the quadrotor altitude and yaw, $z(t)$ and $\psi(t)$, ensured by the controllers (13) and (18). Regarding the latter, the most significative tracking error is 0.5 rad immediately after the initial time.

Figure 8 shows the quadrotor $x$ and $y$ positions ensured by the controllers (30) and (38). The $x$ position achieves its reference value in 3 s, and the $y$ position achieves its reference value in 6 s. The overall controller performance is accurate and fast. Moreover, this performance could be improved using also integral terms in the controllers.
6. Conclusion
In this paper, a controller based on the backstepping technique has been designed for the position and yaw control of a quadrotor helicopter. The quadrotor dynamics have been divided into four subsystems, each with a control input and an output variable. The overall controller leads to satisfactory results. The photogrammetric technique on a scenario with fixed obstacles, due to some trees, has been used to determine the scene geometry and to fix the position reference trajectory. The numerical simulations of the proposed controllers have been implemented using the PSP, ensuring an accurate approximation of the real quadrotor dynamics, and the results show a good performance and effectiveness of the proposed control law. Future work will regard the real implementation of the proposed controller.

Data Availability
The figures, tables, and other data used to support this study are included within the article.

Conflicts of Interest
The authors declare that there are no conflicts of interest regarding the publication of this paper.

Acknowledgments
This work was partially supported by the European Project ECSEL–JURIA–2018 “Comp4Drones” and by the Project “Coordination of Autonomous Unmanned Vehicles for Highly Complex Performances,” Executive Program of Scientific and Technological Agreement between Italy (Ministry of Foreign Affairs and International Cooperation, Italy) and Mexico (Mexican International Cooperation Agency for the Development), SAAP3.

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