Incremental Twisting Fault Tolerant Control for Hypersonic Vehicles With Partial Model Knowledge

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Abstract—A passive fault tolerant control scheme is proposed for the full reentry trajectory tracking of a hypersonic vehicle in the presence of modeling uncertainties, external disturbances, and actuator faults. To achieve this goal, the attitude error dynamics with relative degree two is formulated first by ignoring the nonlinearities induced by the translational motions. Then, a multivariable twisting controller is developed as a benchmark to ensure the precise tracking task. Theoretical analysis with the Lyapunov method proves that the attitude tracking error and its first-order derivative can simultaneously converge to the origin exponentially. To depend less on the model knowledge and reduce the system uncertainties, an incremental twisting fault tolerant controller is derived based on the incremental nonlinear dynamic inversion control and the predesigned twisting controller. In this article, it is shown that not only the benefits of both incremental control and twisting control are inherited, but also their side effects are reduced. Notably, the proposed controller is user friendly in that only fixed gains and partial model knowledge are required. Numerical simulations in various cases and comparison studies are conducted to verify the effectiveness of the proposed method.

Index Terms—Actuator faults, fault tolerant control (FTC), hypersonic vehicle, reentry trajectory tracking, twisting control.

I. INTRODUCTION

SAFETY has been playing an important role in the aerospace industry and is demanding for the flight control of hypersonic vehicles due to its complex structure, civil/military mission, and high cost. The main factors that threaten flight safety are the actuator faults, unmodeled dynamics, and external disturbances [1], [2]. To handle this issue, the fault tolerant control (FTC) has emerged and witnessed a wide range of applications in robots [3], ground vehicles [4], aerial vehicles [5], and spacecraft [6].

Generally, FTC can be classified into two groups: active FTC and passive FTC [7]. The active FTC framework estimates faults via the fault detection and identification (FDI) method. Then, it reconfigures the controller for self-healing utilizing the remaining healthy actuators. For instance, the fault efficiency factor was estimated by an adaptive composite observer in [4] and combined with the fast terminal sliding mode to obtain an active FTC for electric vehicles. In [5], a super twisting observer was utilized to estimate the servo’s stuck fault of an unmanned aerial vehicle (UAV) and an integral FTC law was developed. The active FTC of near-space hypersonic vehicles was studied in [8] via a sliding mode observer-based FDI. In [9], a neural network adaptive FDI process was combined with the nonlinear dynamic inversion (NDI) control to actively compensate for the actuator faults of UAV. To deal with the undetected/missed small actuator faults, an adaptive FDI that only detects but not estimates the fault was integrated within a robust UAV control framework [10]. The total effect of the actuator fault was estimated in [11] to develop an active FTC for spacecraft under input saturation. The key of active FTC is to enhance system robustness relying on the FDI technique that requires more about the model knowledge.

The passive FTC scheme preserves/enhances the system robustness against faults and uncertainties without using FDI. For example, the nonlinear disturbance observer was combined with sliding mode control (SMC) in [2] for the hypersonic vehicle FTC issue. A prescribed-time FTC law was designed in [3] by integrating a barrier Lyapunov function. Based on the efficiency of SMC, a passive FTC law for spacecraft was developed in [6] using the time-varying SMC. Besides, various passive FTC laws were also developed for hypersonic vehicles based on the terminal SMC [12], [13], barrier-Lyapunov function [14], and the singular perturbation-based control [15]. The merit of the
passive FTC is that it can improve computational efficiency compared to the active FTC.

Although the above success has been achieved in FTC, a key problem remains open in this field. Specifically, the FTC will become less robust with either poor FDI performance for the active FTC or inaccurate uncertainty cancellation for the passive FTC. However, if more accurate FDI or uncertainty compensation methods are used, the computational complexity will increase and much more effort will be required to obtain the accurate model knowledge. Notably, the state-of-the-art FTC strategies face the following design difficulties.

1) The strong dependency on the model knowledge.
2) System uncertainty increases with less model knowledge.
3) Greater minimum control gains are required under larger uncertainty magnitude.
4) Severe chattering will occur with larger control gains for SMC-based FTC. It is difficult and contradictory to meet the objectives of reduced uncertainty, less model dependency, smaller control gains, and less chattering phenomenon simultaneously.

To handle such a dilemma, the incremental nonlinear dynamic inversion (INDI) was proposed in [16] and has attracted much attention for the FTC design in the aerospace community [17]–[19]. Recent flight tests [20], [21] have also demonstrated the applicability and efficiency of this method. The main advantage of INDI is that not only the model dependency is reduced, but also the robustness is improved. Recently, the combination of INDI and SMC (named INDI-SMC [18], [19], [22]) that inherits the merits and avoids the defects of both methods has been studied to further improve the robustness of INDI. Motivated by the above mentioned, this article proposes an incremental twisting FTC law that innovatively integrates the INDI and twisting control [23], [24] (a type of second-order SMC) for the full reentry trajectory tracking of hypersonic vehicles. Note that it is more challenging and significant to deal with the full trajectory tracking issue because of the unknown and long-duration flight environment, time-varying aerodynamics, nonlinear dynamics, strong couplings, and especially in the presence of uncertainties, disturbances, and actuator faults. The main contributions of this article are as follows.

1) Contributions to the twisting control: A multivariable twisting controller is proposed for systems with relative degree two. It is shown that the system tracking error and its derivative can exponentially converge to the origin simultaneously.

2) Contributions to the INDI architecture: The INDI is extended with twisting control (named as INDI-TW) to further enhance its robustness. Moreover, the INDI-TW has a simple structure without requiring the design of sliding surfaces.

3) Contributions to the FTC of hypersonic vehicles: The INDI-TW is applied to the FTC for hypersonic vehicles with partial model knowledge under system uncertainties, external disturbances, and actuator faults. Unlike most FTC designs for hypersonic vehicles that only consider a fixed feature point or the longitudinal dynamics, the proposed technique is feasible in tracking a full reentry trajectory with fixed gains.

The rest of this article is organized as follows. The problem is formulated in Section II. The twisting control based on NDI (NDI-TW) and INDI (INDI-TW) is presented in Section III, and the performance of the proposed method is verified in Section IV. Finally, Section V concludes the article.

II. MODEL DESCRIPTION AND PROBLEM FORMULATION

In this section, the robust tracking problem for an aerodynamic surface actuated hypersonic vehicle that behaves bank-to-turn (BTT) maneuvers is formulated under external disturbances, model uncertainties, and actuator faults.

A. Nonlinear Model for Hypersonic Vehicles

The nonlinear six-degree-of-freedom model for hypersonic vehicles can be presented as [25]–[27]

\[
\dot{h} = V \sin \gamma \\
\dot{\phi} = V \cos \gamma \sin \theta / [(r_e + h) \cos \theta] \\
\dot{\theta} = V \cos \gamma \cos \theta / (r_e + h) \\
\dot{V} = \omega_e^2 (r_e + h) \cos \theta (\sin \gamma \cos \theta - \cos \gamma \sin \theta \cos \vartheta) \\
- \frac{D}{m - g} \sin \gamma \\
\dot{\vartheta} = L \sin \sigma / (mV \cos \gamma) + V \cos \gamma \sin \vartheta \tan \theta / (r_e + h) \\
+ \omega_e^2 (r_e + h) \sin \theta \cos \vartheta \sin \vartheta / (V \cos \gamma) \\
- 2 \omega_e (\tan \gamma \cos \theta \cos \vartheta \sin \sigma) \\
\dot{\gamma} = L \cos \sigma / (mV) - g \cos \gamma / V + V \cos \gamma / (r_e + h) \\
+ 2 \omega_e \cos \theta \sin \vartheta \\
+ \omega_e^2 (r_e + h) \cos \theta (\cos \gamma \cos \theta + \sin \gamma \sin \theta \cos \vartheta) / V \\
\] (6)

where \(h\) is the flight altitude, \(V\) is the vehicle’s earth relative velocity, \(\gamma\) is the flight path angle, \(\vartheta\) is the heading angle, and \(\sigma\) is the bank angle. \(\phi\) and \(\theta\) are the latitude and longitude, respectively. \(r_e, \omega_e, g\) denote the radius, angular speed, and gravity constant of the Earth, respectively. \(L, D, m\) are the lift force, drag force, and mass of the vehicle, respectively. Note that the kinematics (1)–(6) is based on the zero sideslip angle assumption due to the BTT maneuver requirement.

2) Rotational Equations:

\[
\dot{\alpha} = q - p \cos \alpha \tan \beta + r \sin \alpha \tan \beta + \sin \sigma / \cos \beta [\dot{\theta} \cos \theta \\
- \dot{\vartheta} \sin \theta \sin \gamma + (\dot{\gamma} + \omega_e) (\cos \vartheta \cos \theta \sin \gamma - \sin \vartheta \cos \gamma)] \\
- \cos \sigma / \cos \beta [\dot{\gamma} - \dot{\vartheta} \cos \theta - (\dot{\theta} + \omega_e) \cos \vartheta \sin \theta \\
- \sin \vartheta \cos \gamma]) + \sin \sigma [\dot{\gamma} - \dot{\vartheta} \cos \theta \\
+ (\dot{\theta} + \omega_e) \cos \vartheta \sin \theta \cos \gamma \\
+ \dot{\vartheta} \sin \theta \sin \gamma - (\dot{\theta} + \omega_e) \cos \vartheta \cos \gamma] / V \\
\] (7)

\[
\dot{\beta} = p \sin \alpha + \cos \sigma [\dot{\theta} \cos \gamma - \dot{\vartheta} \sin \theta \sin \gamma - (\dot{\theta} \\
+ \omega_e) (\cos \vartheta \cos \theta \sin \gamma \\
- \sin \vartheta \cos \gamma)] + \sin \sigma [\dot{\gamma} - \dot{\vartheta} \cos \theta \\
+ (\dot{\theta} + \omega_e) \cos \vartheta \sin \theta \cos \gamma \\
+ (\dot{\theta} + \omega_e) \cos \vartheta \sin \theta \sin \gamma + r \cos \alpha] / V \\
\] (8)
\[
\sigma = -p \cos \alpha \cos \beta - q \sin \beta - r \sin \alpha \cos \beta + \dot{\alpha} \sin \beta - \dot{\theta} \sin \gamma \\
- \phi \sin \theta \cos \gamma + (\dot{\theta} + \omega_r) (\cos \phi \cos \theta \cos \gamma + \sin \phi \sin \gamma)
\]
(9)

\[
\dot{p} = a_1 M_r + a_2 M_y + a_3 pq + a_4 qr
\]
(10)

\[
\dot{q} = a_3 M_p + a_6 (r^2 - p^2) + a_7 pr
\]
(11)

\[
\dot{r} = a_2 M_r - a_3 qr + a_8 M_y + a_9 pq
\]
(12)

where \([a_1, a_2, a_3, a_4, a_5, a_6, a_7, a_8, a_9] = [1, I_{xx}, I_{zz}, -I_{xx}] / I_{yy} \) and \([a_1, a_2, a_3, a_4, a_5, a_6, a_7, a_8, a_9] = [I_{xx}, I_{yy}, I_{zz}, (I_{yy} - I_{zz})I_{xx}, I_{zz}, (I_{xx} - I_{yy})I_{zz}, I_{zz} + I_{xx}] / (I_{xx}I_{zz} - I_{xx}^2).

\(\alpha\) is the angle of attack (AOA), \(\beta\) is the sideslip angle. \(M_r, M_p, M_y\) are the roll, pitch, and yaw moments, \(p, q, r\) are the corresponding roll, pitch, and yaw angular rates, respectively. \(I_{ij}(i, j = x, y, z)\) denotes the moment of inertia.

**B. Aerodynamic Force and Moment Model**

The aerodynamic lift and drag forces are presented as

\[
\begin{bmatrix}
L \\
D
\end{bmatrix} = \begin{bmatrix}
\bar{q} C_L \left( M \alpha, \alpha, \delta_a, \delta_e, \delta_r \right) S_{ref} \\
\bar{q} C_D \left( M \alpha, \alpha, \delta_a, \delta_e, \delta_r \right) S_{ref}
\end{bmatrix}
\]

where \(\bar{q} = 0.5 \rho V^2\) is the dynamic pressure, \(\rho\) is the atmospheric density, \(S_{ref}\) is the reference area. \(C_L, C_D\) are the lift and drag coefficients related to the Mach number \((Ma)\), AOA, and fin deflections \((\delta_a, \delta_e, \delta_r)\).

The roll, pitch, and yaw moments are computed by

\[
\begin{bmatrix}
M_r \\
M_p \\
M_y
\end{bmatrix} = \begin{bmatrix}
\bar{q} m_r \left( M \alpha, \alpha, \delta_a, \delta_e, \delta_r \right) S_{ref} L_{ref} \\
\bar{q} m_p \left( M \alpha, \alpha, \delta_a, \delta_e, \delta_r \right) S_{ref} L_{ref} \\
\bar{q} m_y \left( M \alpha, \alpha, \delta_a, \delta_e, \delta_r \right) S_{ref} L_{ref}
\end{bmatrix}
\]

where \(L_{ref}\) is the reference length, \(m_r, m_p, m_y\) are the roll, pitch, and yaw moment coefficients related to the Mach number, AOA, sideslip angle, and fin deflections.

**C. Attitude Error Dynamics With Uncertainties, Disturbances, and Actuator Faults**

To design a proper attitude controller such that the hypersonic vehicle can precisely track the reference trajectory generated by (1)–(6), the attitude dynamics is presented in a compact form from (7)–(12) as follows:

\[
\begin{cases}
\dot{\Psi} = T \omega \\
\dot{\omega} = -I^{-1} \Gamma I \omega + I^{-1} M
\end{cases}
\]

where \(\Psi = [\alpha, \beta, \sigma]^T, \omega = [p, q, r]^T, M = [M_r, M_p, M_y]^T, \)

\[
T = \begin{bmatrix}
-\cos \alpha \tan \beta & 1 & \sin \alpha \tan \beta \\
\sin \alpha & 0 & \cos \alpha \\
-\cos \alpha \cos \beta - \sin \beta - \sin \alpha \cos \beta & -\cos \alpha \sin \beta + \sin \alpha \cos \beta
\end{bmatrix}
\]

\[
I = \begin{bmatrix}
I_{xx} & I_{xy} & -I_{xz} \\
-I_{yx} & I_{yy} & -I_{yz} \\
-I_{zx} & -I_{zy} & I_{zz}
\end{bmatrix}, \quad \Gamma = \begin{bmatrix} 0 & -r & q \\ r & 0 & -p \\ -q & p & 0 \end{bmatrix}.
\]

Note that (13) neglects the translational part and the Earth’s angular speed in (7)–(9). Such manipulation is reasonable since the translational motion is much slower than the rotational motion for hypersonic vehicles [24]. In practice, to attain the specific control surface deflections, the aerodynamic moment vector \(M\) is further presented as

\[
M = PCu
\]
(14)

where \(u = [\delta_1, \delta_2, \delta_3, \ldots, \delta_p]^T, p \geq 3\) is the actual control surface deflection vector, \(C \in \mathbb{R}^{3p \times p}\) is the control allocation matrix whose elements are \(c_{ij} \in [0, 1] (i = 1, 2, 3, j = 1, 2, 3, \ldots, p)\), and

\[
P = q S_{ref} L_{ref} \begin{bmatrix}
m_{ra} & m_{re} & m_{rr} \\
m_{pa} & m_{pe} & m_{pr} \\
m_{ya} & m_{ye} & m_{yr}
\end{bmatrix}
\]

where \(m_{ij}(i = r, p, y, j = \alpha, \beta, \sigma)\) denote the coefficients related to the partial differential terms of \(m_i\) with respect to \(\delta_a, \delta_e, \delta_r\).

Considering the reference command as \(\Psi_c = [\alpha_c, \beta_c, \sigma_c]^T\) and defining the attitude tracking error as \(e_{\Psi} = \Psi - \Psi_c\), the attitude error dynamics can be obtained from (13) and (14) as

\[
\begin{cases}
\dot{\Psi} = e_{\Psi} \\
\dot{e}_{\Psi} = A + Bu - \dot{\Psi}_c + \Omega
\end{cases}
\]
(15)

where \(e_{\Psi} = T \omega - \dot{\Psi}_c, \quad \dot{\Psi}_c = -TI^{-1} \Gamma I \omega, \quad B = TI^{-1} PC, \quad \Omega = T \omega\).

**Remark 1:** \(\alpha_c, \beta_c, \sigma_c\) are obtained by optimizing (1)–(6) with terminal and process constraints. \(\beta_c\) is set to zero due to the BTT maneuver. Tracking of these angles is a common objective in hypersonic vehicle control problems [13], [24]–[27].

**Assumption 1:** The reference command \(\Psi_c\) is bounded, twice differentiable and smooth, with bounded \(\dot{\Psi}_c\) and \(\ddot{\Psi}_c\).

To further consider the practical situation, the modeling uncertainties, external disturbances, and actuator faults are involved in the second equation of (15). Then, (15) becomes

\[
\begin{cases}
\dot{\Psi} = e_{\Psi} \\
\dot{e}_{\Psi} = A + Bu - \dot{\Psi}_c + \Omega
\end{cases}
\]
(16)

where \(\Omega\) is the lumped external disturbance vector, and

\[
A = \bar{A} + (A_F - \bar{A}) \delta(t) + \Delta A
\]
(17)

\[
B = \bar{B} + (B_F - \bar{B}) \delta(t) + \Delta B
\]
(18)

\[
\Omega = \bar{\Omega} + d
\]
(19)

where \(\Delta A, \Delta B\) are unmodeled dynamics, \(A_F, B_F\) are versions of \(\bar{A}, \bar{B}\) under actuator faults, \(d\) is the external disturbance, \(\delta(t)\) is a function that triggers the fault at \(t = t_f\) and is described as

\[
\delta(t) = \begin{cases} 0, & \text{if } t < t_f \\ 1, & \text{if } t \geq t_f \end{cases}
\]
(20)

**Assumption 2:** The lumped disturbance \(\Omega\) is bounded.

**D. Control Objectives**

1) Stabilize the attitude error dynamics (16) such that \(e_{\Psi} \rightarrow 0\), i.e., the following aims should be achieved

\[
\alpha \rightarrow \alpha_c, \beta \rightarrow \beta_c, \sigma \rightarrow \sigma_c.
\]
2) Design the control law \( u \) using partial model knowledge, i.e., the uncertainties \( \Delta A, \Delta B \), fault information \( \Delta f(t), \delta(t) \), disturbance \( \Omega \), and even the nominal system dynamics \( A, B \), are unknown and will not be compensated via the estimation tools.

### III. INDI-TW Control Scheme

To achieve the control objectives presented in Section II.D, the NDI-TW is first designed and analyzed as a benchmark. Then, the INDI-TW is proposed and its stability is proved. The comparison between NDI-TW and INDI-TW is analyzed.

#### A. NDI-TW Controller Design and Analysis

For the attitude error dynamics (16) with relative degree two, a multivariable twisting controller inspired by [28] can be designed based on the NDI control structure as follows:

\[
\begin{align*}
\mathbf{u}_{\text{ndi-tw}} &= \mathbf{B}^+ \left( -\mathbf{A} + \mathbf{\Psi}_c \right) + \mathbf{B}^+ \left( -k_1 \left\| \mathbf{\xi}_c \right\| - k_2 \left\| \mathbf{\xi}_{eq} \right\| - k_3 \mathbf{\xi}_c - k_4 \mathbf{\xi}_{eq} \right) \\
&= \mathbf{B}^+ \left( -k_1 \frac{\mathbf{\xi}_c}{\left\| \mathbf{\xi}_c \right\|} - k_2 \frac{\mathbf{\xi}_{eq}}{\left\| \mathbf{\xi}_{eq} \right\|} - k_3 \mathbf{\xi}_c - k_4 \mathbf{\xi}_{eq} \right) \\
&= \mathbf{B}^+ \left( -k_1 \frac{\mathbf{\xi}_c}{\left\| \mathbf{\xi}_c \right\|} - k_2 \frac{\mathbf{\xi}_{eq}}{\left\| \mathbf{\xi}_{eq} \right\|} - k_3 \mathbf{\xi}_c - k_4 \mathbf{\xi}_{eq} \right)
\end{align*}
\]

where \( k_1, k_2, k_3, k_4 \) are positive gains to be designed, the pseudo-inverse matrix \( \mathbf{B}^+ = \mathbf{B}^+ (\mathbf{B} \mathbf{B}^T)^{-1} \) and satisfies \( \mathbf{B} \mathbf{B}^T = \mathbf{I} \).

Substituting (21) into (16) leads to

\[
\begin{align*}
\dot{\mathbf{\xi}}_c &= \mathbf{A} + \mathbf{B} \left( \mathbf{B}^+ \mathbf{u}_{eq} + \mathbf{B}^+ \mathbf{u}_{tw} \right) - \mathbf{\Psi}_c + \Omega \\
&= \mathbf{u}_{tw} + \mathbf{A} - \mathbf{\Psi}_c + \mathbf{\Psi}_c + \mathbf{u}_{tw} - \mathbf{u}_{tw} \\text{ (22)}
\end{align*}
\]

Equation (22) can be rewritten as

\[
\begin{align*}
\dot{\mathbf{\xi}}_c &= \dot{\mathbf{\xi}}_{eq} + \mathbf{u}_{tw} + \mathbf{u}_{eq} - \mathbf{u}_{tw} \\text{ (23)}
\end{align*}
\]

where

\[
\mathbf{u}_{eq} = \mathbf{A} - \mathbf{\Psi}_c + \mathbf{B} \left( \mathbf{B}^+ \mathbf{u}_{eq} + \mathbf{B}^+ \mathbf{u}_{tw} \right)
\]

represents the uncertainty residual existing in the error dynamics (16) under the NDI-TW controller (21).

**Assumption 3:** The residual \( \mathbf{u}_{eq} \) presented in (24) is bounded and satisfies \( \left\| \mathbf{u}_{eq} \right\| \leq \chi \) with \( \chi \) a positive scalar.

**Remark 2:** Assumption 3 is also a primary hypothesis in many robust control methods (e.g., [1], [2], [6], [11], [14], [24], [27], to name a few).

**Theorem 1.** For the attitude error dynamics (16) or the transformed (23), the NDI-TW controller (21) can guarantee the simultaneous convergence of \( \mathbf{\xi}_c \) and \( \mathbf{\xi}_{eq} \) exponentially under the condition of (24) for some \( \bar{m} > 1, \varsigma_1 > 0, \varsigma_2 > 0 \).

**Proof:** It will first be shown that the equilibrium \( \mathbf{e}_c = \mathbf{e}_{eq} = 0 \) is a unique point of the closed-loop system (23). Specifically, the error dynamics with \( \mathbf{e}_{eq} = 0 \) can be described as [28]

\[
\begin{align*}
\dot{\mathbf{e}}_c &= 0 \\
\dot{\mathbf{e}}_{eq} &= -k_1 \frac{\mathbf{e}_{eq}}{\left\| \mathbf{e}_{eq} \right\|} - k_3 \mathbf{e}_c + (\bar{\nu}_2 - \bar{\nu}_3) \mathbf{v}
\end{align*}
\]

where \( -k_2 \leq \bar{\nu}_1 \leq k_2, -\chi \leq \bar{\nu}_2 \leq \chi \), \( \mathbf{v} = [1, 1]^T \). It can be seen that \( \dot{\mathbf{e}}_c \neq 0 \) unless \( \mathbf{e}_c = 0 \) because \( k_1 > k_2 + \chi \) as per (25).

\[
\dot{W} (\mathbf{e}_c, \mathbf{e}_{eq}) = c_1 \left\| \mathbf{e}_c \right\|^2 + c_2 \mathbf{e}_c^T \mathbf{e}_{eq} + c_3 \left\| \mathbf{e}_{eq} \right\|^2 + c_4 \left\| \mathbf{e}_c \right\|
\]

where \( c_1, c_2, c_3, c_4 > 0 \) are constants. Reformulating (27) as

\[
W (\mathbf{e}_c, \mathbf{e}_{eq}) = \mathbf{e}_c^T \mathbf{Q} \mathbf{e}_c + c_4 \left\| \mathbf{e}_c \right\|
\]

where

\[
\mathbf{Q} = \begin{bmatrix} c_1 & c_2/2 \\ c_2/2 & c_3 \end{bmatrix}
\]

Apparent, \( W (\mathbf{e}_c, \mathbf{e}_{eq}) \) is positive definite (except for the origin) if \( 4c_1c_3 > c_2^2 \). Note that the Lyapunov function is not differentiable at \( \mathbf{e}_c = 0 \). However, the error dynamics with the condition \( (0, \mathbf{e}_{eq}) \) can be checked as follows: [28]

\[
W (\mathbf{e}_c, \mathbf{e}_{eq}) = 2c_1 \mathbf{e}_c^T \mathbf{e}_{eq} + c_2 \left\| \mathbf{e}_{eq} \right\|^2
\]

which is positive definite because \( k_1c_2 > k_2c_2 \). It can be seen that \( \dot{W} (\mathbf{e}_c, \mathbf{e}_{eq}) \neq 0 \) for nonzero \( \mathbf{e}_c \) which means the closed-loop system (23) will not stay on \( e_c = 0 \) for \( e_{eq} \neq 0 \). Then, taking the derivative of (28) along the system (23) leads to

\[
\dot{W} (\mathbf{e}_c, \mathbf{e}_{eq}) = -k_1c_2 \left\| \mathbf{e}_c \right\| - k_2c_2 \mathbf{e}_c^T \mathbf{e}_{eq} + c_4 \left\| \mathbf{e}_c \right\|
\]

To proceed, (29) is rewritten as

\[
\dot{W} (\mathbf{e}_c, \mathbf{e}_{eq}) = -(W_1 + W_2)
\]

(30)
where

\[
W_1 = c_2 e_\Psi^T \begin{pmatrix} k_1 e_\Psi ||e_\Psi|| + k_2 e_\omega ||e_\omega|| - \varepsilon_{\text{ndi-tw}} \end{pmatrix} - c_4 e_\Psi^T e_\omega \\
+ 2c_3 e_\omega^T \begin{pmatrix} k_2 e_\omega ||e_\omega|| + k_1 e_\Psi ||e_\Psi|| - \varepsilon_{\text{ndi-tw}} \end{pmatrix}
\]

\[
W_2 = k_3 c_2 ||e_\Psi||^2 + (2k_4 c_3 - c_2) ||e_\Psi||^2 \\
+ (2k_3 c_3 + k_4 c_2 - c_1) e_\Psi^T e_\omega.
\]

(31)

Note that \(\dot{W}(e_\Psi, e_\omega) < 0\) if \(W_1 > 0\) and \(W_2 > 0\). From (31), it can be obtained that

\[
W_1 \geq c_2 ||e_\Psi|| (k_1 - k_2 - \chi) + 2c_3 ||e_\omega|| (k_2 - \chi)
+ 2k_1 c_3 e_\Psi^T e_\Psi/||e_\Psi|| - c_4 e_\Psi^T e_\omega/||e_\omega||.
\]

Let \(c_4\) be selected as \(c_4 = 2k_1 c_3 / \tilde{m}\) with \(\tilde{m} > 1\), then

\[
W_1 \geq c_2 ||e_\Psi|| (k_1 - k_2 - \chi) + 2c_3 ||e_\omega|| (k_2 - k_1 - k_1 / \tilde{m} - \chi)
\]

(33)

It can be seen from (33) that \(W_1 > 0\) (except for the origin) if

\[
\begin{cases}
  k_1 > k_2 + \chi \\
  k_2 > k_1 (1 - 1 / \tilde{m}) + \chi
\end{cases}
\]

which can be met by the selection of \((\tilde{m})\).

For the positivity of \(W_2\), \(c_1, c_2, c_3\) are chosen such that

\[
\begin{align*}
&c_2 = 2\tilde{m} c_3 \\
&2\tilde{m} c_1 = 2k_3 c_3 + \tilde{m} k_4 c_2
\end{align*}
\]

(34)

Therefore, it can be derived that

\[
\frac{c_2^2}{4c_1 c_3} = \frac{\tilde{m}^2 c_3}{c_1} = \frac{\tilde{m}^3}{k_3 + k_4 \tilde{m}}
\]

(35)

Obviously, \(4c_1 c_3 > c_2^2\) for the positivity of (28) will not be violated with \(k_4 > \tilde{m}\). It can be deduced from (32) and (34) that

\[
W_2 = 2\tilde{m} k_3 c_3 ||e_\Psi||^2 + (2k_4 c_3 - 2\tilde{m} c_1) ||e_\Psi||^2
+ 2k_1 c_3 (1 - 1 / \tilde{m}) e_\Psi^T e_\omega = c_3 \xi^T R \xi
\]

where

\[
R = \begin{bmatrix} 2\tilde{m} k_3 k_1 (1 - 1 / \tilde{m}) \\
2 (k_3 - \tilde{m}) \end{bmatrix}
\]

(36)

Combining (30) with (36) yields

\[
\dot{W}(e_\Psi, e_\omega) \leq - \left( c_2 \xi^T Q \xi + c_3 \lambda_{\min} \{ R \} \xi^2 \right)
\]

\[
\leq - G_1 (\xi^T Q \xi + ||e_\Psi||^2)
\]

(37)

where \(G_1 = \min \{ c_2 \xi_1, c_3 \lambda_{\min} \{ R \} \}\). Recall (27) and consider that \(\lambda_{\min} \{ Q \} ||\xi||^2 \leq \xi^T Q \xi \leq \lambda_{\max} \{ Q \} ||\xi||^2\), it can be derived that

\[
W(e_\Psi, e_\omega) \leq \lambda_{\max} \{ Q \} ||\xi||^2 + c_4 ||e_\Psi|| \leq G_2 \left( ||\xi||^2 + ||e_\Psi|| \right)
\]

(38)

where \(G_2 = \max \{ \lambda_{\max} \{ Q \}, c_4 \}\). Hence, it can be obtained from (37) and (38) that

\[
\dot{W}(e_\Psi, e_\omega) \leq - \frac{G_1}{G_2} W(e_\Psi, e_\omega).
\]

Therefore, the system is globally exponentially stable. In other words, the error dynamics \(e_\Psi\) and \(e_\omega\) can simultaneously converge to the origin exponentially. This completes the proof.

Remark 3: Although the NDI-TW controller has the property of cancelling perturbations, there exist several conservativeness.

1) The strong dependency on the model knowledge [both \(\hat{A}\) and \(\hat{B}\) are required in (21)].

2) The residual \(e_{\text{ndi-tw}}\) increases when the knowledge of \(\hat{A}, \hat{B}\) becomes worse [see (24)].

3) The larger the residual’s absolute values, the greater minimum values of the control gains \(k_1, k_2\) are required as per (25). Therefore, it is challenging and contradictory to reduce the model dependency and uncertainty residual simultaneously for the NDI-TW controller. This issue will be mitigated by designing the INDI-TW controller in the next subsection.

B. INDI-TW Controller Design and Analysis

This section derives an INDI-TW controller that exploits the incremental control concept [18] in which the angular acceleration measurement and partial model dynamics are used. To begin with, the second equation of (16) is rewritten by considering \(A, B\) are functions of the system state (denoted as \(x\)) and fault trigger \(\delta(t)\) as

\[
\dot{e}_\omega = A(x, \delta) + B(x, \delta) u - \tilde{\Psi}_\omega + \Omega.
\]

(39)

Then, the incremental dynamics for \(e_\omega\) is derived via taking the first-order Taylor series expansion of (39) around the instant \(t - \Delta t\) (described by the subscript 0, \(\Delta t\) is the sampling interval) [18] as follows:

\[
\dot{e}_\omega = \dot{e}_{\omega 0} + \frac{\partial [A(x, \delta) + B(x, \delta) u]}{\partial x} \bigg|_0 \Delta x + B(x_0, \delta_0) \Delta u
\]

\[- \Delta \tilde{\Psi}_\omega + \frac{\partial [A(x, \delta) + B(x, \delta) u]}{\partial \delta} \bigg|_0 \Delta \delta + \Delta \Omega + O(\Delta x^2)
\]

(40)

where \(\dot{e}_{\omega 0}\) is the estimated/measured derivative of the error dynamics \(e_{\omega}\) in the latest step, \(\Delta x = x - x_0\) and \(\Delta u = u - u_0\) denote the increments of \(x\) and \(u\) from \(t - \Delta t\) to \(t\). \(\Delta \delta = \delta - \delta_0\) and \(\Delta \Omega = \Omega - \Omega_0\) depict the fault trigger changes and the variations of the disturbances in \(\Delta t\), respectively. \(O(\Delta x^2)\) denotes the higher-order remaining function.

Considering (17), (18), and (40), (16) is rewritten as

\[
\dot{e}_\Psi = e_{\omega 0} + B_0 (x_0, \delta_0) \Delta u + \Delta \tilde{\Psi}_\omega + \Delta \Omega + \lambda (x, \delta, \Delta t)
\]

(41)

where \(B_0 = B(x_0, \delta_0)\), and

\[
\lambda (x, \delta, \Delta t) = \lambda_1 + \lambda_2 + \lambda_3
\]

(42)
\[ \lambda_1 = \frac{\partial \left[ \bar{A} + \Delta A + (\bar{B} + \Delta B) u \right]}{\partial x} \bigg|_0 \Delta x + O \left( \Delta x^2 \right) \]  
(43)

\[ \lambda_2 = \delta_0 \frac{\partial \left[ (A_F - \bar{A}) + (B_F - \bar{B}) u \right]}{\partial x} \bigg|_0 \Delta x \]  
(44)

\[ \lambda_3 = \Delta \delta \left[ (A_F - \bar{A}) + (B_F - \bar{B}) u \right] \bigg|_0. \]  
(45)

Assumption 4: [19]. The partial derivatives of \( A \) and \( B \) with respect to \( x \) are bounded for any order.

Assumption 5: [19]. The term \( \lambda_F \) in (45) is bounded for \( t_F, t_F + \Delta t \).

For the system (41) with relative degree two, the INDI-TW controller is designed as

\[ \Delta u_{\text{indi-tw}} = \bar{B}_0^+ \left( -\dot{e}_{x0} + \bar{\Psi}_x + u_{\text{twi}} \right) \]  
(46)

where \( u_{\text{twi}} \) is the same as \( u_{\text{tw}} \) defined in (21).

Remark 4: Note that \( \dot{e}_{x0} \) can be obtained by angular sensors because \( \dot{e}_x = T\dot{\omega} - \bar{\Psi}_x + \bar{T}\dot{\omega} \), where \( \bar{T}\dot{\omega} = \bar{\Omega} \) has been included in the lumped disturbance (19) whose effects will be analyzed later. For some cases where \( \dot{\omega} \) is not available by sensors, the sampled outputs can be utilized for its estimation [19]. This is not the focus of this work and will not be discussed in detail.

Remark 5: If \( k_1, k_2 \) in \( u_{\text{eqi}} \) are set to zero, then (46) changes from the INDI-TW controller to the INDI controller with reduced robustness.

Remark 6: The unit vectors in \( u_{\text{tw}} \) and \( u_{\text{twi}} \) are modified as \( \bar{e}_x = e_x / \left\| e_x \right\| + \bar{e}_x \), \( \bar{e}_{x0} = e_{x0} / \left\| e_{x0} \right\| + \bar{e}_{x0} \) to avoid chattering in practical implementation. Increasing \( \bar{\omega} \) will reduce the system robustness, while reducing \( \bar{\omega} \) will increase the possibility of chattering, and vice versa.

The whole control command can be presented as

\[ u_{\text{indi-tw}} = u_0 + \Delta u_{\text{indi-tw}} \]

where \( u_0 \) is the actuator deflection measured at the latest step \( t - \Delta t \). Note that if \( u_0 \) cannot be directly measured, it can be also estimated online [29].

Substituting (46) into (41) leads to

\[ \begin{cases} \dot{e}_x = e_x \\ \dot{e}_{x0} = u_{\text{twi}} + e_{\text{indi-tw}} \end{cases} \]  
(47)

where

\[ e_{\text{indi-tw}} = (B_0\bar{B}_0^+ - I) (u_{\text{eqi}} + u_{\text{twi}}) + \Delta \Omega + \lambda (x, \delta, \Delta t). \]  
(48)

Then, the following theorem derived from [18] can be obtained.

Theorem 2: Under the Assumptions 1, 2, 4, and 5, the system residual \( e_{\text{indi-tw}} \) presented in (48) is ultimately bounded if the condition \( \| I - B_0\bar{B}_0^+ \| \leq c < 1 \) is met for all \( t \).

Proof: From (47), the value of \( \dot{e}_x \) for the previous step can be presented as

\[ \dot{e}_{x0} = u_{\text{twi0}} + e_{\text{indi-tw0}}. \]  
(49)

Substituting \( u_{eqj} = -\dot{e}_{x0} + \bar{\Psi}_x + u_{\text{twi0}} \) into (48) and combining with (49), it can be derived that

\[ e_{\text{indi-tw}} = \left( I - B_0\bar{B}_0^+ \right) (e_{\text{indi-tw0}} - \bar{\Psi}_x - \Delta u_{\text{twi}}) + \Delta \Omega + \lambda (x, \delta, \Delta t) \]  
(50)

where \( \Delta u_{\text{twi}} = u_{\text{twi}} - u_{\text{twi0}} \). For a sufficiently small \( \Delta t \), it is assumed that \( \Delta u_{\text{twi}} \) is bounded by \( \Theta_i \) under continuous modification of \( u_{\text{twi}} \) in implementation. It can be noted from Assumptions 1 and 2 that \( \bar{\Psi}_x \) and \( \Delta \Omega \) are bounded and their bounds are denoted as \( \Theta_2, \Theta_3 \), respectively. For \( \lambda (x, \delta, \Delta t) \), it can be known that \( \lambda_1 \) and \( \lambda_2 \) are bounded under Assumption 4 because \( \| \Delta x \| \rightarrow 0 \) with a sufficiently small \( \Delta t \) [19], [22]. Meanwhile, \( \lambda_3 \) is also bounded under Assumption 5 because \( \Delta \delta = 0 \) only for \( [t_F, t_F + \Delta t] \) and \( \Delta \delta = 0 \) for any other time. Consequently, the boundedness of \( \lambda (x, \delta, \Delta t) \) is ensured and denoted as \( \Theta_4 \). Rewriting (50) recursively as:

\[ e_{\text{indi-tw}}(k) = \left( I - B_0\bar{B}_0^+ \right) e_{\text{indi-tw}}(k - 1) + \Delta \Omega(k) + \lambda(k) \]

Then, with the condition \( \| I - B_0\bar{B}_0^+ \| \leq c < 1 \), it can be deduced from [18] and [19] that

\[ \| e_{\text{indi-tw}}(k) \| \leq c^k \| e_{\text{indi-tw}}(0) \| + \frac{\left[ c(\Theta_1 + \Theta_2) + (\Theta_3 + \Theta_4) \right] k}{1-c} \sum_{i=1}^{k} c^{k-i} \leq c^k \| e_{\text{indi-tw}}(0) \| + \frac{c(\Theta_1 + \Theta_2) + (\Theta_3 + \Theta_4)}{1-c}. \]  
(51)

Since \( c^k \rightarrow 0 \) as \( k \rightarrow +\infty \) for \( 0 \leq c < 1 \), and \( \| e_{\text{indi-tw}}(0) \| \) is finite in practice, then (51) becomes

\[ \| e_{\text{indi-tw}}(k) \| \leq \bar{c} = \frac{c(\Theta_1 + \Theta_2) + (\Theta_3 + \Theta_4)}{1-c}. \]

This completes the proof.

Remark 7: The bound of the residual \( e_{\text{indi-tw}} \) is related to \( \Delta t \), and its boundary can be further reduced with higher sampling frequency [18], [19], [22].

Remark 8: The condition \( \| I - B_0\bar{B}_0^+ \| \leq c < 1 \) requires \( B_0\bar{B}_0^+ \) to be diagonally dominant, which is commonly used in flight control [18]. Note that it is a condition relaxing Assumption 3. For some cases where the condition is invalid, Assumption 3 should be also made for \( e_{\text{indi-tw}} \) to ensure the system stability. Particularly, \( \| I - B_0\bar{B}_0^+ \| = 0 \) if \( \bar{B}_0 \) ideally estimates \( B_0 \). It gives a future research direction in the accurate identification of \( B_0 \) such that \( \| e_{\text{indi-tw}} \| \) can quickly decay with faster convergence for \( c^k \rightarrow 0 \).

Corollary 1: For the attitude error dynamics (16) or the transformed incremental dynamics (41) under uncertainties, faults, and disturbances (17)–(19), the proposed INDI-TW controller (46) can guarantee the simultaneous exponential convergence of \( e_x \) and \( e_{x0} \) under the bounded \( \| e_{\text{indi-tw}} \| \) if the control
C. Comparisons Between NDI-TW and INDI-TW

The conservativeness in the NDI-TW controller can reduce the magnitude of $\bar{\epsilon}_{ndi-tw}$ and $\chi$ can be further reduced with smaller $\lambda$. As per (43) and (44), $\|\lambda_1\|$ and $\|\lambda_1 + \lambda_2\|$ are negligible for a small $\Delta t$. Meanwhile, if $\Omega \neq 0$ and $u_{ndi-tw} \neq 0$, there exists a sufficiently small $\Delta t$ such that $|\Delta t\Omega| < \|\Omega\|, |\Delta u_{ndi-tw}| < \|u_{ndi-tw}||$ [18], [19] under continuous modification of $u_{tw}$ and $u_{tw}$ in implementation. Hence, a smaller bound of $\epsilon_{ndi-tw}$ can be attained for a small $\Delta t$ than $\epsilon_{ndi-tw}$ except for $t \in [t_F, t_{F+}]$, during which $\epsilon_{ndi-tw}$ and $\epsilon_{ndi-tw}$ are comparable. As for the guidance of choosing $\Delta t$, one can refer to [30].

**Remark 10:** The gains $k_1, k_2$ for both NDI-TW and INDI-TW controllers are related to $\chi$ and $\chi$, respectively. It is alternative to use gain adaption methods to update $k_1, k_2$ such as the adaptive twisting controller in [24]. However, it is not the focus in this work and does not affect the comparative analysis for $\epsilon_{ndi-tw}$ and $\epsilon_{ndi-tw}$.

To sum up, the INDI-TW-based FTC for hypersonic vehicles has the following theoretical and practical significance. 1) The dependency on part of the model knowledge (dynamics in $A$) is removed for controller design and uncertainty induction. 2) Small and fixed gains can be attained for full trajectory tracking due to reduced residual magnitude, which improves robustness and attenuates chattering. 3) Fault tolerance and tracking accuracy can be enhanced with higher sampling frequency.

## IV. NUMERICAL VERIFICATION

The performance of the NDI-TW and INDI-TW is validated for the tracking of a hypersonic vehicle’s reentry trajectory. The vehicle model in [26] is adopted. Specifically, the mass, reference area, reference length, and the moments of inertia are 1200 kg, 0.446 m², 0.98 m, $I_{zz} = 100$ kg·m², $I_{yy} = 5600$ kg·m², $I_{zz} = 5700$ kg·m², respectively. The aerodynamic coefficients in [26] are used for the trajectory design and attitude tracking. The initial conditions are shown in Table I. The attitude commands are obtained by optimizing the reentry trajectory via the Gauss Pseudo-Spectral Method to meet the following constraints:

$h_f = 30$ km, $V_f = 2300$ m/s, $\gamma_f = -1.9\degree$, $\theta_f = 55\degree$,

$$[\alpha_{min}, \alpha_{max}] = [5\degree, 15\degree],$$

$$[\sigma_{min}, \sigma_{max}] = [-60\degree, 60\degree],$$

$q \leq 120$ kpa,

$$n \leq 5 \text{ g}, \bar{I} \leq 1300 \text{ KWh/m}^2$$
where \( n \) is the total overload, \( \dot{I} \) is the heating flux density, the subscript “f” corresponds to final states. The objective is to meet the minimum flight time, which is 297.24 s. The control gains for NDI-TW are selected by the following steps.

1) \( \bar{m} = 1.1, k_3 = 2.2, k_4 = 2 \) are selected to meet \( \bar{m} > 1, k_3 > 0, k_4 > \bar{m} \), and the condition in (26).
2) \( \varsigma_1 = \varsigma_2 = 0.01 \) are chosen to satisfy \( \varsigma_1 > 0, \varsigma_2 > 0 \).
3) \( k_1 \geq 0.68 \) and \( k_2 \geq 0.37 \) are calculated from (25) by estimating a conservative \( \varsigma = 0.3 \).

Then, \( k_1 = 0.8 \) and \( k_2 = 0.7 \) can be easily tuned to meet acceptable dynamic responses. The control gains for INDI-TW are selected the same as NDI-TW for a fair comparison. The INDI controller with \( \{k_1, k_2, k_3, k_4\} = [0, 0, 2, 2, 2] \) is also verified. The nominal \( A \) and/or \( \dot{B} \) are used for all controllers. The sampling frequency is 200 Hz, \( \varsigma \omega = 0.1 \) is selected to make a balance between chattering issue and system robustness according to Remark 6. The actuator dynamics is modeled by a second-order transfer function [25] with the natural frequency of 20 Hz and the damping ratio of 0.7. Besides, the actuator’s position and rate are limited by \( |\delta| \leq 20^\circ \) and \( |\dot{\delta}| \leq 100^\circ/\text{s} \).

A. Verifications Under Nominal Condition

The first set of simulations is conducted under the nominal condition. As can be seen from Fig. 2, all the controllers can precisely track the attitude commands during the whole reentry phase. Only small errors arise during the AOA transition intervals. Both NDI-TW and INDI-TW have almost the same performance, while INDI creates slightly larger errors because it’s less robust against the strong nonlinearities and couplings in real flight dynamics. Fig. 3 shows that smooth and normal control commands are generated by all controllers. Moreover, it can be observed from Fig. 4 that the vehicle tracks the reference trajectory precisely with small deviations.

B. Verifications Under External Disturbances

To evaluate the robustness of each controller, the external disturbances are chosen as \( d = \bar{d} \times [1 + \sin(\pi t/100) + \sin(\pi t/200), 1 + \sin(\pi t/100) + \cos(\pi t/200), 1 + \cos(\pi t/100) + \sin(\pi t/200)]^T \). \( \bar{d} \) is the magnitude to be tested. It can be seen from Fig. 5 that the tracking performance is barely influenced compared to Fig. 2.

### Table I
INITIAL FLIGHT CONDITIONS OF THE HYPERSONIC VEHICLE

| Variables          | Values     | Variables | Values |
|-------------------|------------|-----------|--------|
| Height            | 60 km      | AOA       | 5 deg  |
| Speed             | 4000 m/s   | Sideslip Angle | 5 deg  |
| Flight Path Angle | 0 deg      | Bank Angle | -13.4 deg |
| Longitude         | 45 deg     | Pitch Rate | 1 deg/s |
| Latitude          | 30 deg     | Roll Rate | 1 deg/s |
| Heading Angle     | 0 deg      | Yaw Rate  | 1 deg/s |

Fig. 2. Attitude responses and tracking errors under the nominal condition.

Fig. 3. Control commands under the nominal condition.

Fig. 4. Flight trajectory responses under the nominal condition.
when \( \bar{d} \) is set to 0.01. However, the NDI-TW and INDI create larger tracking errors than INDI-TW when \( \bar{d} \) increases to 0.1, as shown in Fig. 6. Note that the INDI and INDI-TW can still ensure stable tracking in this case, while the NDI-TW cannot.

It is worth noting that the tracking errors can be reduced by NDI-TW if higher control gains are set. To this end, the attitude tracking errors and control commands under NDI-TW with \( K \) and \( 2K \) (\( K \) denotes all the control gains) are compared in Fig. 7. It can be observed that the tracking errors are reduced by high gains, but the chattering arises, which is undesirable in practice.

### C. Verifications Under Disturbances and Actuator Faults

The external disturbances with \( \bar{d} = 0.1 \) and actuator faults are simultaneously considered to validate the fault-tolerance and robustness of the three controllers. Four actuators are considered as \( u = [\delta_{le}, \delta_{ur}, \delta_{re}, \delta_{lr}]^T \) whose components are left elevon, upper rudder, right elevon, and lower rudder, respectively. The allocation matrix satisfying \( C u = [\delta_a, \delta_r, \delta_e]^T \) is \( C = [0.25, 0.25, -0.25, -0.25; 0, 0.5, 0, 0.5; 0.5, 0, 0.5, 0]^T \). The left elevon \( \delta_{le} \) suddenly lost its effectiveness by 50% during the reentry phase at \( t = 10 \) s. This fault will cause severe perturbations in the roll and pitch moments. Therefore, the tracking of AOA and bank angle is disturbed, and thus the sideslip angle will be affected in turn due to the strong couplings among them.

From Figs. 8 and 9, it can be noted that the NDI-TW is incapable of handling the fault and the states leave away from the references with large errors. The other two controllers succeed in the tracking mission, and the INDI-TW has better performance and smaller errors than INDI. The attitudes controlled by INDI and INDI-TW diverge first after the fault occurs but are quickly driven back to their stable points. This is because the fault case is recovered using the remaining healthy actuators, as can be seen in Fig. 10. It can be observed that \( \delta_{le} \) under INDI and INDI-TW is near zero after fault happens such that the vehicle will depend less on the left elevon. At the same time, the other three actuators make immediate responses to the faulty actuator for recovery. As shown in Fig. 11, the norm of the uncertainty residual by NDI-TW is greater than that by INDI-TW, which validates the analysis in Section III-C.

For the sensor-based controller investigated in this article, the control performance under noisy measurements is further demonstrated using the INDI-TW controller. Specifically, Gaussian measurement noise with zero mean and standard deviations of 0.001 rad/s, 0.01 rad/s², and 0.01 rad is considered for the angular rate (\( \omega \)), the latest angular acceleration-related (\( \dot{\omega}_0 \)) term, and the latest actuator (\( u_0 \)), respectively. The simulation...
setting is the same as that described in the beginning of this subsection. In addition, a low-pass filter is employed to smooth the noisy measurement. The attitude and actuator responses are shown in Figs. 12 and 13, respectively. It can be seen that the tracking errors are acceptable, and the tracking performance as well as the fin deflections are not affected much, compared to the results under INDI-TW in Figs. 8 and 10.

V. CONCLUSION

The full reentry trajectory tracking problem for hypersonic vehicles was solved considering modeling uncertainties, external disturbances, and actuator faults. A multivariable twisting controller was proposed to ensure the tracking error and its first-order derivative converge to the origin simultaneously and exponentially. To depend less on the model knowledge, an incremental twisting fault tolerant controller was developed and its closed-loop stability was proved with theoretical guarantees on the boundedness of uncertainty residuals. Simulations validated that both the INDI-TW and INDI can feasibly track the full trajectory with fixed control gains even under disturbances, uncertainties, and actuator faults, while NDI-TW failed the same mission. Comparison studies showed that the INDI-TW has better robustness and fault tolerance compared to the NDI-TW and INDI. Although the tracking errors can be reduced by NDI-TW with high gains, undesirable chattering is introduced. The future research will focus on the input saturation and estimation of angular acceleration.

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